WHAT IS CLAIMED:

1. A gas turbine stationary blade comprising:

a stationary blade section provided therein with a passage for cooling air;

an inner shroud for supporting the stationary blade section on the side of a discharge port of the cooling air; and

a plurality of segments each of which includes at least one of the stationary blades and at least one of the inner shrouds, and which are coupled to one another in an annular form, wherein the segments are assembled in such a manner that a side end face of the inner shroud is connected to that of another inner shroud of an adjacent segment by using a bolt joint,

wherein at least one flow passage is pulled out from the discharge port of the cooling air of the stationary blade section, and the flow passage is introduced to a corner section on the side of a front edge of the inner shroud and is extended rearward along a side edge of the inner shroud.

2. The gas turbine stationary blade according to claim 1, wherein the flow passage meanders in a front edge corner section of the inner shroud.

3. The gas turbine stationary blade according to claim 1, wherein a turbulator is installed on the floor section or a ceiling section of the widened flow passage.

4. The gas turbine stationary blade according to claim 1, wherein an acicular fin is formed from the floor section to the ceiling section of the widened flow passage.

5. The gas turbine stationary blade according to claim 1, wherein film cooling holes are formed by being pulled out from the flow passage and are provided on a front edge corner section of the inner shroud.

6. A gas turbine stationary blade comprising:

a plurality of segments coupled to one another in an annular form, each of said plurality of segments including;

a first inner shroud and a second inner shroud fixed to each other side by side, wherein each of the first and second inner shrouds has a first edge and a second edge, and a first side edge and a second side edge, the first edge being on an upstream side and the second edge being on a downstream side with respect to gas flow, wherein the first and second inner shrouds being fixed to each other on the first side edges thereof, with the second side edges thereof outside, and wherein each of the first and second inner shrouds has a corner section defined by the first edge and the second side edge;

a first stationary blade section and a second stationary blade section, each of which is provided therein with a passage for cooling air and has a discharge port for the cooling air on one end, wherein the first and second stationary blade sections are supported by the first and second inner shrouds, respectively, on the end of the blade section where the discharge port is provided;

wherein the segments are assembled in such a manner that a side end face of one of the inner shrouds in one segment is connected to that of another of the inner shrouds in an adjacent segment by using a bolt joint; and

wherein each of the first and second inner shrouds is provided therein with a flow passage that communicates with the discharge port and extends in the corner section and along the second side edge toward the second edge.

7. The gas turbine stationary blade according to claim 6, wherein the flow passage has a meandering portion in the corner section.

8. The gas turbine stationary blade according to claim 6, wherein a turbulator is installed on the floor section or a ceiling section of the widened flow passage.

9. The gas turbine stationary blade according to claim 6, wherein an acicular fin is formed from the floor section to the ceiling section of the widened flow passage.

10. The gas turbine stationary blade according to claim 6, wherein the corner section of the inner shroud is provided with film cooling holes that communicate with the flow passage.

11. The gas turbine stationary blade according to claim 6, wherein each of the first and second inner shrouds is provided therein with another flow passage that communicates with the passage for cooling gas and extends along the first side edge.